

CRYOGENIC TELESCOPE ON THE  
INFRARED ASTRONOMICAL SATELLITE (IRAS)

F. J. Low<sup>a</sup>, C. A. Beichman<sup>b</sup>, F. C. Gillett<sup>c</sup>, J. R. Houck<sup>d</sup>,  
G. Neugebauer<sup>e</sup>, D. E. Langford<sup>f</sup>, R. G. Walker<sup>g</sup>, R. H. White<sup>h</sup>

Abstract

The Infrared Astronomical Satellite (IRAS) has completed an unbiased all-sky survey at wavelengths from 10 to 100  $\mu\text{m}$ . The design and performance of the focal plane array is described with emphasis on in-orbit measurements of the sensitivity and stability. In the four broad spectral bands centered at 12, 25, 60 and 100  $\mu\text{m}$  the system NEFD values are, in  $\text{Jy}/\text{Hz}^{1/2}$ , 0.03, 0.025, 0.046 and 0.21 respectively (Jansky =  $1\text{E}-26 \text{ w}/\text{m}^2/\text{Hz}$ ). For point sources, a single scan at the survey rate of 3.8 arcmin/sec yields limiting flux densities at the 3 sigma confidence level of 0.36, 0.30, 0.39 and 1.2 Jy. The DC stability of the JFET amplifiers and the excellent off-axis rejection of the telescope permit total flux measurements of extended infrared emission at levels below  $1 \text{ E } 6 \text{ Jy}/\text{sr}$ . Response of the extrinsic silicon and germanium photodetectors to ionizing radiation is described.

Introduction

In June of 1975 scientists and engineers from the United States, The Netherlands, and the United Kingdom formed a team to evaluate the feasibility of performing an all-sky survey at infrared wavelengths from 10 to 100  $\mu\text{m}$  using a cryogenically cooled space borne telescope. The mission, which was approved late in 1975, would build upon experience gained from the Astronomical Netherlands Satellite (ANS) and on the advanced infrared detector and cooled telescope technologies demonstrated by the series of sub-orbital rocket flights carried out by workers at Cornell University (1,2,3), the Naval Research Laboratory (4,5), and the Air Force Geophysical Laboratory (6). The scientific requirements which led to the design of the survey focal plane are discussed below. The prime contractor chosen by NASA was Ball Aerospace Systems Division with responsibility for the focal plane assembly given to Rockwell International.

The first version of the IRAS array was delivered to NASA's Ames Research Center early in 1981 for testing prior to integration into the telescope system. It became apparent at that time that the noise level defining the NEP of the array was that of the MOSFET preamplifiers, considerably higher than the desired Johnson noise of the load resistors and too high to meet the scientific objectives. Also the offset voltage of the MOSFETS was unpredictable from one cool-down to another leading to poor DC stability. The susceptibility of MOSFETS to static discharge damage constituted an unacceptable risk to the focal plane.

The array was transferred to the Jet Propulsion Laboratory (JPL) in mid 1981 where it was retrofitted with self-heating JFET preamps produced by Infrared Laboratories, and new detectors for the 100  $\mu\text{m}$  band from Ge:Ga material processed by Cornell University. The new array was integrated into the telescope system early in 1982 after extensive testing at JPL.

- a Steward Observatory, University of Arizona, Tucson, Arizona 85721
- b JPL, California Institute of Technology, Pasadena, California 91109
- c Kitt Peak National Observatory, Tucson, Arizona 85726
- d CPSR, Cornell University, Ithaca, New York 14853
- e California Institute of Technology, Pasadena, California 91125
- f JPL, California Institute of Technology, Pasadena, California 91109
- g Jamieson Science and Engineering, Inc., Palo Alto, California 94306
- h JPL, California Institute of Technology, Pasadena, California 91109

This paper discusses the design and performance of the IRAS focal plane array which was launched on 25 January 1983, and is performing satisfactorily in orbit.

Scientific Objectives of IRAS  
and Conceptual Design of the Focal Plane

A number of important scientific objectives had to be considered in the design of the IRAS focal plane.<sup>(7)</sup> From a scientific point of view it was desirable to survey the sky over a broad spectral range in a number of contiguous bands at high sensitivity. Concerns about the availability of high sensitivity detectors, high quality filter and field lens materials as well as the technical complexity of the assembly resulted in the adoption of the four wavelength bands from 8 to 120 microns. The more complex issues of the detailed focal plane layout, field aperture size and shape were decided on the basis of the several rather specific requirements on the principal goal of the mission --- an unbiased all-sky survey.

An all-sky survey is scientifically useful only if it is both complete and reliable. A complete catalog is one in which all of the sources in the sky down to some limiting flux level are included. In practice it is impossible to insure that every source is included without exception. A statistical completeness requirement of 98% was placed on the system. A reliable catalog is one which does not contain any false - nonexistent - sources. The statistical requirement on the catalog is that it be 99.8% reliable. A higher requirement was set on reliability because false sources would probably have rather "interesting properties" such as peculiar colors or rapidly varying intensity. If such sources were present in the catalog in large numbers, then considerable time would be required to separate them from the genuinely interesting discoveries. The above requirements were waived in the area immediately surrounding the galactic plane because of its expected high source density.

Translating the completeness and reliability requirements into hardware specifications was a difficult process. We consider now the various features of the focal plane to see how they contribute to the dual requirements of completeness and reliability. As can be seen from the layout shown in Figure 3, each color band is represented by two double rows of detectors. Each double row is called a module and contains either seven or eight detectors. Each module is approximately 1/2 degree long in the cross-scan direction as projected on the sky. The in-scan width of the apertures was set at several times the diffraction size for the module's central wavelength. The cross scan length of the apertures in each band was determined by the expected source density on the sky and the requirement that false sources not be generated by the accidental combination of two or more real sources. Theoretical models of the expected source density as a function of flux level and position on the sky were developed to support this activity. The edges of two adjacent apertures overlap sufficiently to insure that a source cannot pass undetected between them. Field lenses are used behind the apertures to reimage the exit pupil of the telescope on the entrance aperture in front of each detector. In this way the photometric response of the individual apertures is smoothed and smaller detectors could be used. The design of a single module is optimized for completeness while at the same time avoiding confusion.

The second module in each color band is included for redundancy and to reduce the number of false sources. A large number of effects can produce source-like pulses in the output data stream, including residual tails from deglitched charged particle events, contamination from the satellite, and sightings of nearby space debris. A true astronomical source should produce one or two output deflections as its image passes over the first module of a given color band; a few seconds later it should produce a second deflection(s) as its image passes over the second module. The time delay between these detections depends on the focal plane geometry, the telescope plate scale and the scan rate. All of these factors are known at the time of the data reduction and can be used to confirm real sources and reject false ones. Because the apertures in the second module are shifted in the cross-scan direction relative to the

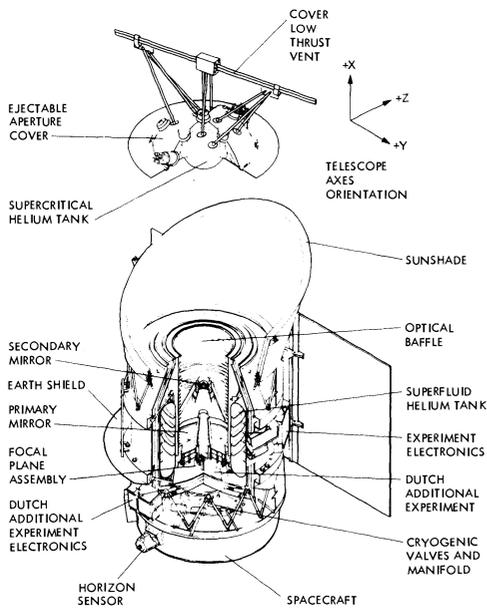


Figure 1. Satellite Configuration

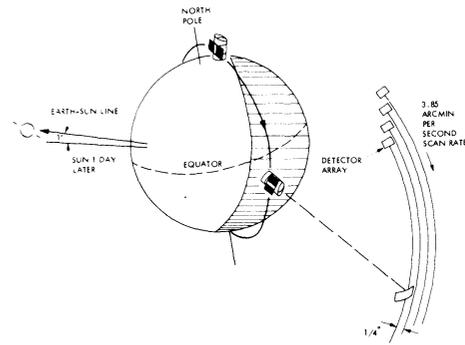


Figure 2. IRAS Orbit Geometry

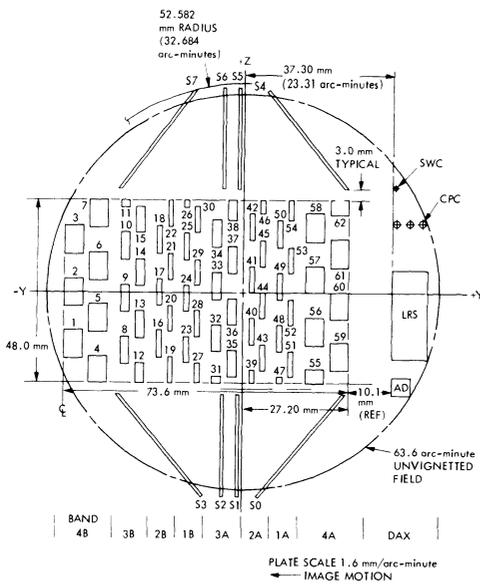


Figure 3. Layout of focal plane showing detector bands and channel numbers.

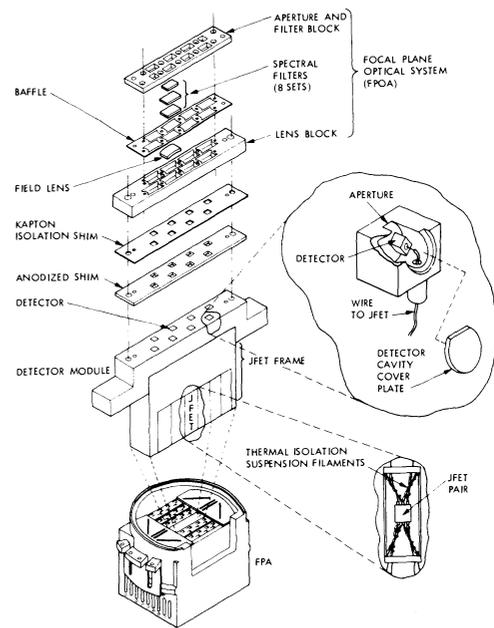


Figure 4. Detector subarray module showing filters, optics, detectors and JFET amplifiers.

apertures in the first module, there is an improvement in the cross-scan determination of source position. An increase in reliability at the expense of some completeness is achieved by rescanning the same area of sky on the next orbit (103 minutes later). A further increase in reliability is gained by repeating the entire process a week to ten days later. The prelaunch models indicated that the combined goals of completeness and reliability could be achieved using the focal plane shown as designed and the procedures described above. The actual flight data indicate that these goals are being met over all of the sky with the exception of the vicinity of the galactic plane.

As the result of balloon observations (8), it was realized early in the design phase that the sky, especially near the galactic plane, would contain complex extended emission in the far infrared and that this would require a stable DC coupled readout of the infrared detectors. Although there is no absolute intensity reference, the DC stability of the detectors was designed to permit total flux mapping with stable electronic baselines. Figure 9 shows scans over the same portion of sky by six different detectors on consecutive orbits. The repeatability of IRAS measurements of weak extended features is obvious.

### Instrument Design

#### General Design

The design of the IRAS system as a whole, including the superfluid helium cooling system developed by BASD, is described elsewhere (9). The satellite configuration shown in Figure 1 and the IRAS orbital geometry, Figure 2, are shown for reference.

#### Design of the Focal Plane Assembly

The Focal Plane Assembly (FPA) contains the infrared and visible detectors and associated field optics. It comprises 62 infrared channels and eight visible channels. The infrared channels are divided into eight modules, two for each color band. Each module contains either seven or eight detectors. Figure 3 shows the layout of the FPA. Infrared sources scan across the FPA parallel to the "Y" axis in all observational modes. Figure 4 shows a cutaway of a single channel of an FPA module. The infrared radiation passes through the spectral filters and is focused on to the detector element by the field lens. The photoconductive detector element responds to the infrared by altering its electrical resistance. A pair of junction field effect transistors (JFET) amplifies the photocurrent from each detector. The visible wavelength channels are of a similar construction using silicon diode detectors and a single metal oxide surface field effect transistor (MOSFET) preamplifier per detector. The visible wavelength detectors are placed in a "V" arrangement to provide two-axis spacecraft attitude information during star crossings.

Table 1

Focal plane assembly	
Detector operating temperature	3.5 K
JFET operating temperature	70 to 80 K
MOSFET operating temperature	3.5 K
Number of detectors	62 infrared, 8 visible
Power dissipation	14 mW
Construction	Modular:
	8 Infrared subarrays
	2 Visible subarrays
Detector material	
Band 1	Si: As
Band 2	Si: Sb

Band 3	Ge: Ga
Band 4	Ge: Ga
Visible	Si
<b>Filter Material</b>	
Band 1	Ba F <sub>2</sub> , Ge
Band 2	Si
Band 3	Sapphire, KRS-5
Band 4	Sapphire, CaF <sub>2</sub> , KCl
Visible	BG-38
<b>Field lens material</b>	
Band 1	Ge
Band 2	Si
Band 3	Ge
Band 4	Ge
Visible	N/A

Table 1 lists the relevant parameters of the design. Figure 5 shows the relative spectral response of the four infrared filter/detector combinations which define the wavelength response of the system.

#### Design of the Electronic Data System

An important aspect of the IRAS design is the analog and digital data system which amplifies and processes the photocurrent provided by the detectors. A key element in this design is the self-heated cryogenically cooled JFET preamplifier described by Low (10). The flow of data through amplification, an on-board deglitcher to reduce the effect of ionizing particle events, analog filtering and, finally, digitization is depicted in Figure 6.

#### Performance of the Detection System

The IRAS focal plane and its related systems were tested prior to launch. Table 2 summarizes the results of these tests and their analysis. It is, of course, the inflight performance of the various components in the system which is of great interest for the users of the IRAS and for designers of future systems. Table 3 gives the results of in-orbit tests under operational conditions.

A comparison of these two sets of test results show that the in-orbit performance equals or exceeds expected performance based on pre-launch tests. Clearly, the astronomers have been provided an excellent infrared detection system for study of the infrared sky.

#### Ionization Radiation Effects

It was found prior to launch but after the design was completed that the infrared detectors used in the focal plane would suffer large changes in noise and responsivity as the result of passage through the intense radiation concentrated in the South Atlantic Anomaly (SAA). The "bias boost" strategy anneals the radiation effects in the detectors by raising the bias voltage so that the detectors "break down" during SAA passages. Figures 9 and 10 show in-flight test results on the performance of the bias boost strategy. A dramatic improvement is seen in the behavior of the 60 and 100 um Ge:Ga detectors.

Other aspects of the particle environment such as cosmic ray hit rates and performance of the on board "deglitcher" are very much as anticipated.

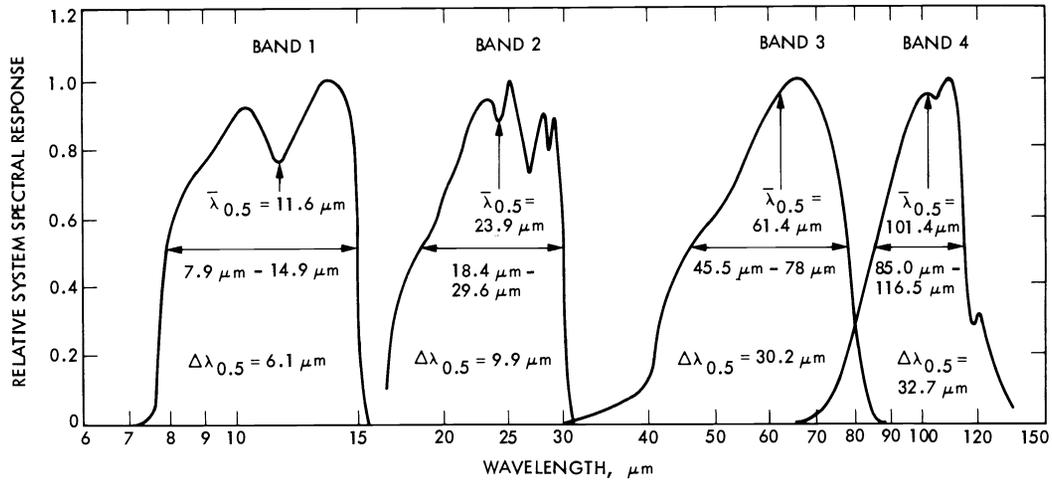


Figure 5. Relative system spectral response curves for the I.R. bands.

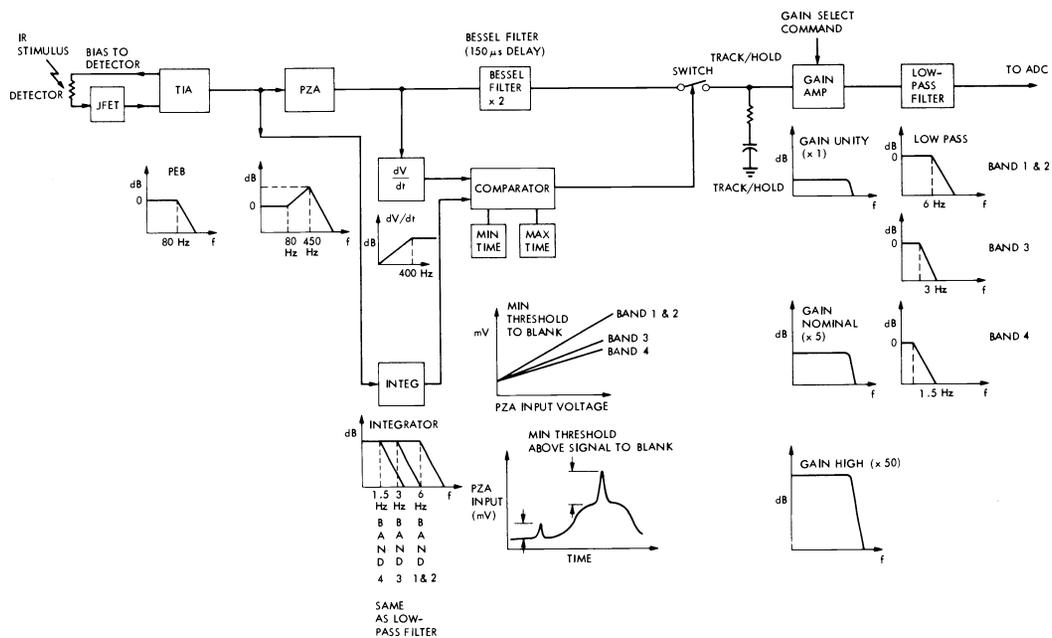


Figure 6. Infrared data flow.

TABLE 2

	Band 1	Band 2	Band 3	Band 4
Spectral response				
Center wavelength	11.8 um	24.4 um	61.6um	101.2 um
Effective bandwidth	5.7 um	9.8 um	31.0 um	35.6 um
Mean inband transmission(%)	0.83	0.81	0.28	0.15
Out-of-band leaks				
shortward leak	$<1.7 \times 10^{-5}$	$<4.5 \times 10^{-5}$	$<2.3 \times 10^{-4}$	$<9.3 \times 10^{-2}$
longward leak	$<2.8 \times 10^{-6}$	$<1.5 \times 10^{-6}$	$<1.5 \times 10^{-4}$	$<5.4 \times 10^{-2}$
Limiting sensitivity (NEFD)				
Survey mode(Jy)	0.12	0.07	0.20	0.30
Deep Sky Mode (Jy)	0.04	0.02	0.07	0.10
Optical quality				
Diameter of 80% encircled energy	25 arc sec	25 arc sec	50 arc sec	100 arc sec
Surface reflectivity	0.96	0.96	0.96	0.96
Aperture efficiency	0.88	0.85	0.87	0.92
Off-axis source rejection				
A( $\odot$ ) for Jupiter @ 5 deg	$1.4 \times 10^0$	$0.26 \times 10^0$	$1.8 \times 10^{-2}$	$1.1 \times 10^{-2}$
A( $\odot$ ) for Moon @ 24 deg	$1.0 \times 10^{-5}$	$8.0 \times 10^{-6}$	$4.0 \times 10^{-6}$	$5.0 \times 10^{-6}$
A( $\odot$ ) for Sun @ 60 deg	$5.0 \times 10^{-8}$	$1.0 \times 10^{-7}$	$2.0 \times 10^{-7}$	$1.0 \times 10^{-7}$
A( $\odot$ ) for Earth limb @88 deg	$2.0 \times 10^{-8}$	$9.0 \times 10^{-9}$	$4.0 \times 10^{-9}$	$4.0 \times 10^{-9}$
Photometric error goal	<15%	<15%	<15%	<15%

\* 1 Jansky =  $10^{-26}$  W/m<sup>2</sup> Hz

Table 3 In-Flight performance of IRAS Focal Plane

Band	Optical Efficiency	System Responsivity (amp/watt)	Detector Responsivity (amp/watt)	NEP (watt/Hz <sup>1/2</sup> )	NEFD (Jy/Hz <sup>1/2</sup> )
I	0.47	0.115	0.35	$3 \times 10^{-16}$	0.03
II	0.44	0.53	1.7	0.6	0.025
III	0.55	0.34	1.0	1.0	0.046
IV	0.16	0.24	1.8	0.6	0.21

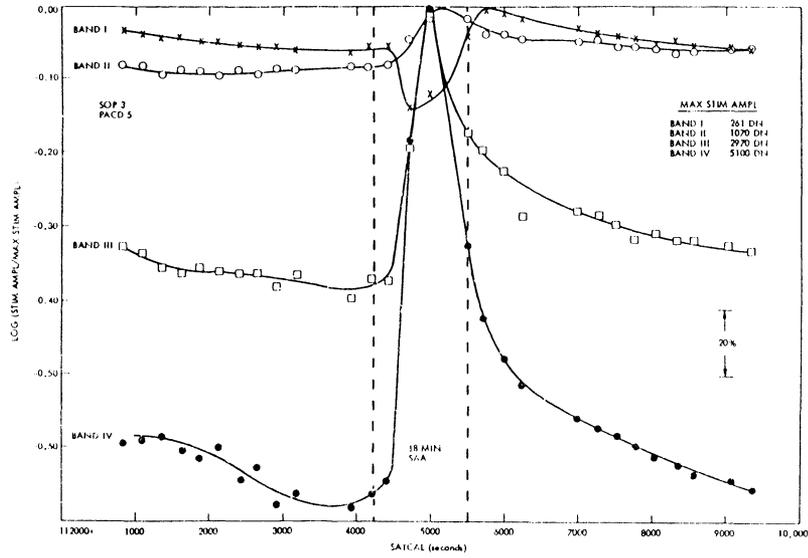


Figure 7. SAA effect on detector response without bias boost.

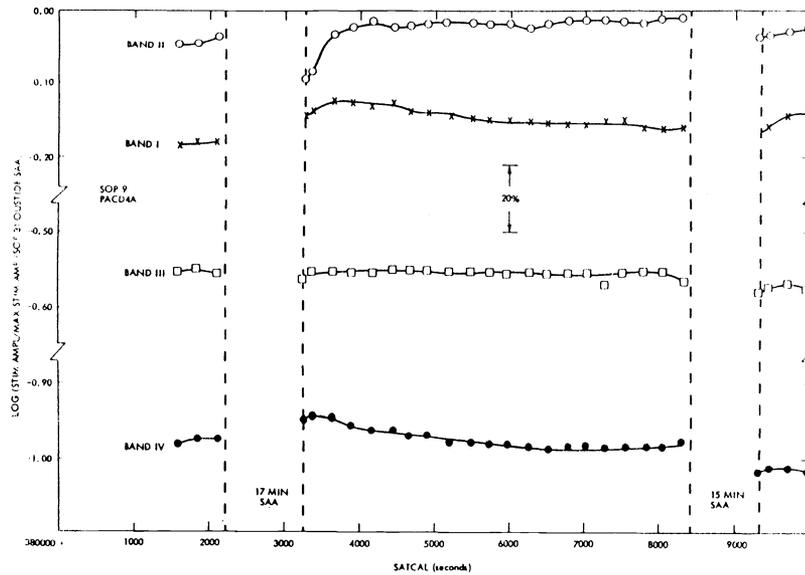


Figure 8. SAA effect on detector response with bias boost.

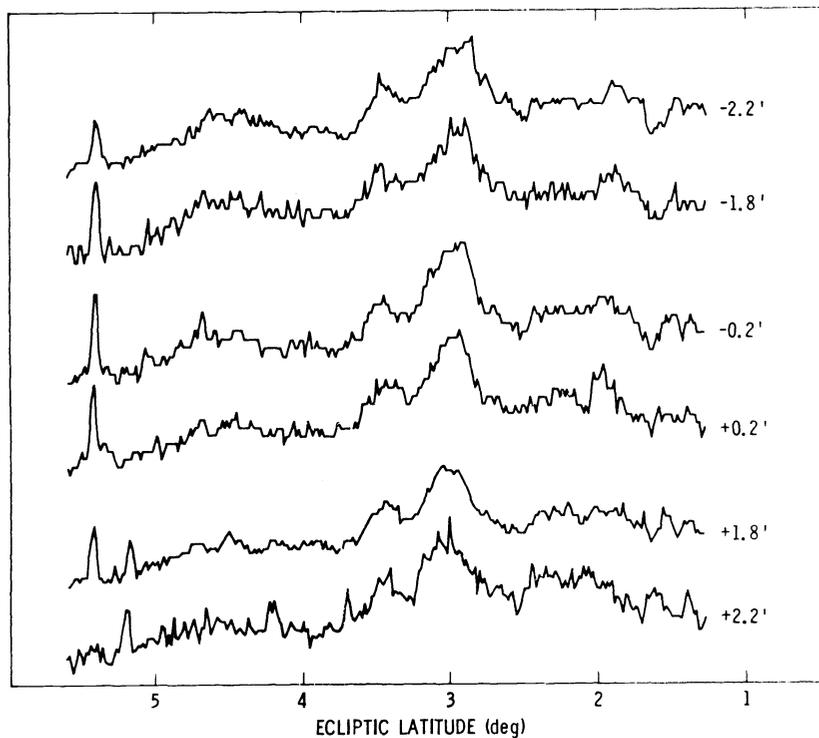


Figure 9. Analog outputs for three pairs of 100 um detectors showing dc repeatability and stability. Each pair of detectors view the same sky but on consecutive orbits.

#### Acknowledgements

The IRAS system performance stands as a lasting tribute to the dedication and skills of its designers and builders.

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#### References

1. Harwit, M. O., Nutt, D. P., Shivanandan, K. and Zajac, B. J., Applied Optics, v1 5, no 11, 1966.
2. Harwit, M. O, Houck, J. R., Fuhrman, K., Applied Optics, vol 8, 473, 1969.
3. Soifer, B. T, Houck, J. R., and Harwit, M., Ap. J. 168, L73, 1971.
4. McNutt, D. P. Shivanandan, K., and Feldman, P. D., Applied Optics, vol 8, no. 11, 1969
5. McNutt, D. P., Shivanandan, K., Daehler, M. D., and Feldman, P. D., Applied Optics, vol 14, no. 5, 1975.
6. Walker, R. G. and Price, S. D., AFGL Four Color Infrared Sky Survey, AFGL-TR-76-0208, 17 Sept 1976.
7. McCreight, C. R., Walker, R. G., and Witteborn, F. C., Proc. of SPIE, vol 132, pp 66, 1978.
8. Nishimura, T., Low, J. J., and Kurtz, R. F. 1980, Ap. J., vol. 239, L101.
9. Irace, W. and Rosing, D. 1983, J. Brit. Interplanetary Soc., vol. 36, 27.
10. Low, F. J. 1981, Proc. SPIE, vol. 280, 56.